SUBORBITAL ANALYSIS: THE A-to-B **PROBLEM IN PLANETARY SCIENCE**

TIME OF FLIGHT (TOF): CORRELATING EJECTA & STREWN TO SOURCE

Derivations Within This SUBORBITAL ANALYSIS Are Based On The Simplified Two-Body Model Where The Satellite Is Assumed To Be Massless. System Mass Is Concentrated At The Center Of The Central Body, Which Is Also The Coordinate Origin Of The Body-Centered Inertial Frame. Higher Order Terms Are Neglected, Such A Planetary Oblateness, Lunar Gravity, Solar Pressure, Electro-Magnetic & Atmospheric Effects,

Basic Suborbital Trajectory: A-to-B Chord & Central Angles (Scaled To Earth's Gravity)



Normal and Oblique Plane Views The b-Circle For TOF Calculation Per Kepler's 2nd Law: Constant Area Sweep Rate



The Oblique Plane View Makes The Orbit Into A Circle Of Radius = "b": b Is The Semi-Minor Axis, So That Calculating Swept Area Is Trivial

Infinite Trajectories Exist To Get From A-to-B: One Solution For Each Discrete TOF While B Rotates Through Inertial Space.











Infinite Different A-to-B Trajectories Exist, Each With A Different TOI Value. For A Rotating Planet. This Complicates Analysis.



The Set Of All Solution Trajectories For A Given A-To-B Pair May Be Defined By The A-To-B Launch Solution Helix. This Usef Format Always Has Common Features From Bottom Up:

- ◆ A Base Lea Starting At The Min TOF Solution Trajectory
- ♦ A Minimum KE Point Just Above Min TOF Point
- A Transition Or "Knee" where ΔEL Gives Way To ΔAZ
- An AZ Arc Which May Encircle 1, 2 or No Poles A "Day Later" Point On Approximately The Min TOF AZ

The Min TOF Trajectory Is Defined By A Circular Orbit At Zero Altitude, Smooth Spherical Planet, No Atmosphere. The Launch Solution Helix Is Defined In The Local Topocentric (Earth-Fixed) Frame For Comparison To Lab Test Ejection Patterns



Launch Vectors in Azimuth (AZ), Elevation (EL), and Magnitude Normalized To Earth Escape KE Or EEKE.

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The Perimeter Of The Conical Flow Displaced By The Obstructions. These Features From Analytics, Computation An Hypervelocity Tes Results Up To M Give High Confidence While Interpreting Imprinted Result







Ref [1]

7) The Hypersonic Entrained Aggregate Flow Is Imprinted By Obstructions AND Suborbital Mechanics

REFERENCES

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TIME OF FLIGHT (TOF): CORRELATING EJECTA & STREWN TO SOURC

ecause So Many Trajectories Are Possible For Every A-to-B Pair, We Need Perspective On How Suborbital Analysis May Help Correlate Both <u>Regional</u> And <u>Global</u> Strewn & Ejecta. Two Master Plots Are Very Helpful For Such Perspective, Allowing Quick Reference For Most Related Problems. On The Left, "Iso-TOF" Contours Are Shown Vs. Central Flight Angle & Normalized

emi-Major Axis. On The Right, Central Flight Angle Contours Are Shown Vs. Eccentricity & TOF.



<u>REGIONAL EXAMPLE:</u> Since The <u>Launch Solution Helix</u> Has A Fairly Vertical Leg For Close A To-B Range, Emplaced Principal Clocking Stays Relatively Constant For Elevations Below The Knee Of The Helix, While Emplaced In-Track Length Increases With Elevation. This Pronounced Separation Of Effects Allows Range-&-Radial Source Location For Repetitive Emplaced

Morphometry That Has Systematic Alignment By Geographic Location (i.e. The Carolina Bays).



EMPLACED PRINCIPAL DIRECTIONS Infinite A-to-B Traiectories Yield A Continua O Downrange & Cross-Range Principals At Point B Different EMPLACED PRINCIPALS For Each TOP



EMPLACED PRINCIPALS Shown At Left In Blue & Above In Red & Black. At Launch Elevations Near Or Above The "Knee" Of The Helix, The In-Track Principal Extends Radically (Black) Vs. Below The Knee (Red).

LOBAL EXAMPLE: For Strewn Distribution, Launch Solution Helices Of Each Fall Site May Be Collated In KE-Space For Any Possible Launch Point A, And The Group Compared To Hyperelocity Test Results For Ejection Trends Matching Specific Test Conditions (i.e. Volatiles, etc.)



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